



US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No. 2120-0020
11/30/2007

Electronic Tracking Number

For FAA Use Only

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a))

1. Aircraft	Nationality and Registration Mark N4082S	Serial No. CE-555	
	Make Beech	Model F33A	Series
2. Owner	Name (As shown on registration certificate) Jasvir S Khurana	Address (As shown on registration certificate)	
		Address	
		City	
		Zip	

3. For FAA Use Only

4. Type		5. Unit Identification			
Repair	Alteration	Unit	Make	Model	Serial No.
☐	☒	AIRFRAME		(As described in Item 1 above)	
☐	☐	POWERPLANT			
☐	☐	PROPELLER			
☐	☐	APPLIANCE	Type		
			Manufacturer		

6. Conformity Statement

A. Agency's Name and Address		B. Kind of Agency	
Name SMART Avionics Inc	Address City Zip	☐ U. S. Certified Mechanic	☐ Manufacturer
		☐ Foreign Certified Mechanic	C. Certificate No.
		☒ Certified Repair Station	
		☐ Certified Maintenance Organization	

D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.

Extended range fuel per 14 CFR Part 43 App. B ☐	Signature/Date of Authorized Individual Benjamin M Travis	5-2-19
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7. Approval for Return to Service

Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is ☒ Approved ☐ Rejected

BY	FAA Fit. Standards Inspector	Manufacturer	Maintenance Organization	Persons Approved by Canadian Department of Transport
	FAA Designee ☒	Repair Station	Inspection Authorization	Other (Specify)

Certificate or Designation No.	Signature/Date of Authorized Individual Benjamin M Travis	5-2-19
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4082S

5/2/19

Nationality and Registration Mark

Date

GFC 500 Autopilot and G5 Installation

Removed vacuum driven Attitude Indicator, Vacuum Driven DG, Vacuum Pump, Vacuum regulator, Vacuum Filter and associated plumbing.

Installed Garmin G5 Electronic Attitude Indicator P/N 011-03809-00 in same location. System was installed in accordance with Garmin Installation manual 190-01112-10 Rev 18 and Garmin STC SA01818-WI. Aircraft is approved on the AML STC.

Installed Garmin G5 Electronic HSI P/N 011-03809-00 in same location as removed KI525 HSI Installed Garmin GMU 11 Magnetometer P/N 010-01788-01 in aft fuselage. Installed GAD29 interface adapter P/N 010-01172-01 on behind pilot instrument panel. System was installed in accordance with Garmin installation manual 190-01112-10 rev 18. System installation is approved in accordance with Garmin STC SA01818WI and aircraft is listed on AML.

System was interfaced to the Garmin GNS530W for Navigation information. The G5 and the GMU11 are powered from a 5 amp milspec circuit breaker P/N 7274-2-5 located on the aircraft main DC bus. The GAD 29 is powered by a 5 amp milspec circuit breaker P/N 7274-2-5 located on the aircraft avionics DC bus.

Installed Garmin GFC 500 Autopilot System in accordance with Garmin Installation manual 190-02291-00 rev 9 and Garmin installation addendum 190-02291-21. System installation was previously approved by Garmin STC SA01866WI.

Installed GSA 28 Pitch Servo P/N 011-02927-11 at airframe sta. 207, GSA 28 Roll Servo P/N 011-02927-11 at sta.112, GSA 28 Pitch Trim Servo P/N 011-02927-11 at sta. 207. Installed GMC507 P/N 011-04548-01 in lower left pilot instrument panel. System was interfaced to existing Garmin GNS430W for Navigation information, Pilot G5 Attitude and G5 HSI for course and information.

Installed Garmin GNC255A Nav/COMM P/N 011-02806-00 in #2 position in accordance with Garmin installation manual 190-01182-02 rev H. System was interfaced to existing MD200-306 Ind to display Nav information, the PM6000 Audio panel for audio control and existing VHF Nav and Comm antennas.

System was ground tested and performed as specified with no adverse effects to other aircraft systems.

All workmanship was accomplished in accordance with the guidelines of AC 43.13-1B chapter 7 sections 2 para 7-14 thru para 7-17, section4 para 7-63 and 7-64 chapter 10 section 2 para 10-1 and 10-2. AC43.13-2B chapters 1 para100 para 106 thru para 114, chapter 2 para 202 and 203.

Electrical load does not exceed aircraft load capabilities in accordance with AC43.13-2B chapter 2 para 208.

Aircraft weight and balance was calculated and the Aircraft flight manual was updated to reflect this change.

See attached Instructions for continued airworthiness.

☒ Additional Sheets Are Attached

4. Instructions for Continued Airworthiness

This document is designed as Instructions for Continued Airworthiness in accordance with 14 CFR 23.1529 Appendix G for use by the owner/operator of the airplane to maintain the GFC 500 Automatic Flight Control System installed under STC SA01866WI.

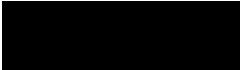
4.1 Airworthiness Limitations

The GFC 500 Autopilot is airworthy when installed, configured, and maintained in accordance with this section.

There are no new or additional Airworthiness Limitations associated with this equipment and/or this installation.

The Airworthiness Limitations section is FAA-approved and specifies maintenance required under §§ 43.16 and 91.403 of Title 14 of the Code of Federal Regulations, unless an alternative program has been FAA-approved.

FAA APPROVED


Robert G. Murray
STC Unit Administrator


3/15/2019
Date