

CESSNA AIRCRAFT COMPANY
MODEL 340
SUPPLEMENTAL INSPECTION DOCUMENT

SUPPLEMENTAL INSPECTION NUMBER: 32-30-00

1. TITLE

Main/Nose Landing Gear Retraction Systems Teardown and Inspection

2. EFFECTIVITY

340-0001 Thru 340A1817

INSPECTION COMPLIANCE

TYPICAL:	INITIAL	10,000 Landings	Or	20 Years
	REPEAT	5000 Landings	Or	10 Years

3. PURPOSE

To inspect for cracks and excessive wear in mechanisms, bushings, bearings, attachment holes in structure, and attaching hardware which could hinder proper rigging and cause gear-down position failures or structural failures.

4. INSPECTION INSTRUCTIONS

- A. Remove all gear assemblies and retraction mechanism parts and hardware from the airplane. Refer to the Service Manual. Refer to Figure 1.
- B. Inspect the wing, nose wheel well, and supporting structures for cracks, corrosion, and elongated attachment holes. Repair or replace, as required.
- C. Inspect all components of the gear and retraction mechanism for cracks, corrosion, and excess wear, and replace with new parts/components where required.
- D. Reinstall all components and rig the system in accordance with the Service Manual.
- E. Inspect 1.19 inch diameter trunnion in accordance with MEB88-5 Revision 2.

5. ACCESS AND DETECTABLE CRACK SIZE

ACCESS/LOCATION	DETECTABLE CRACK SIZE
Refer to the above inspection instructions.	0.25 Inch

6. INSPECTION METHOD

Refer to the procedure above.

7. REPAIR/MODIFICATION

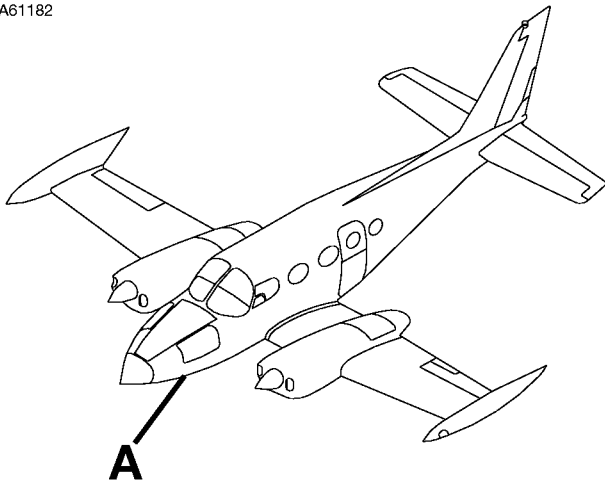
Refer to the procedure above.

8. COMMENTS

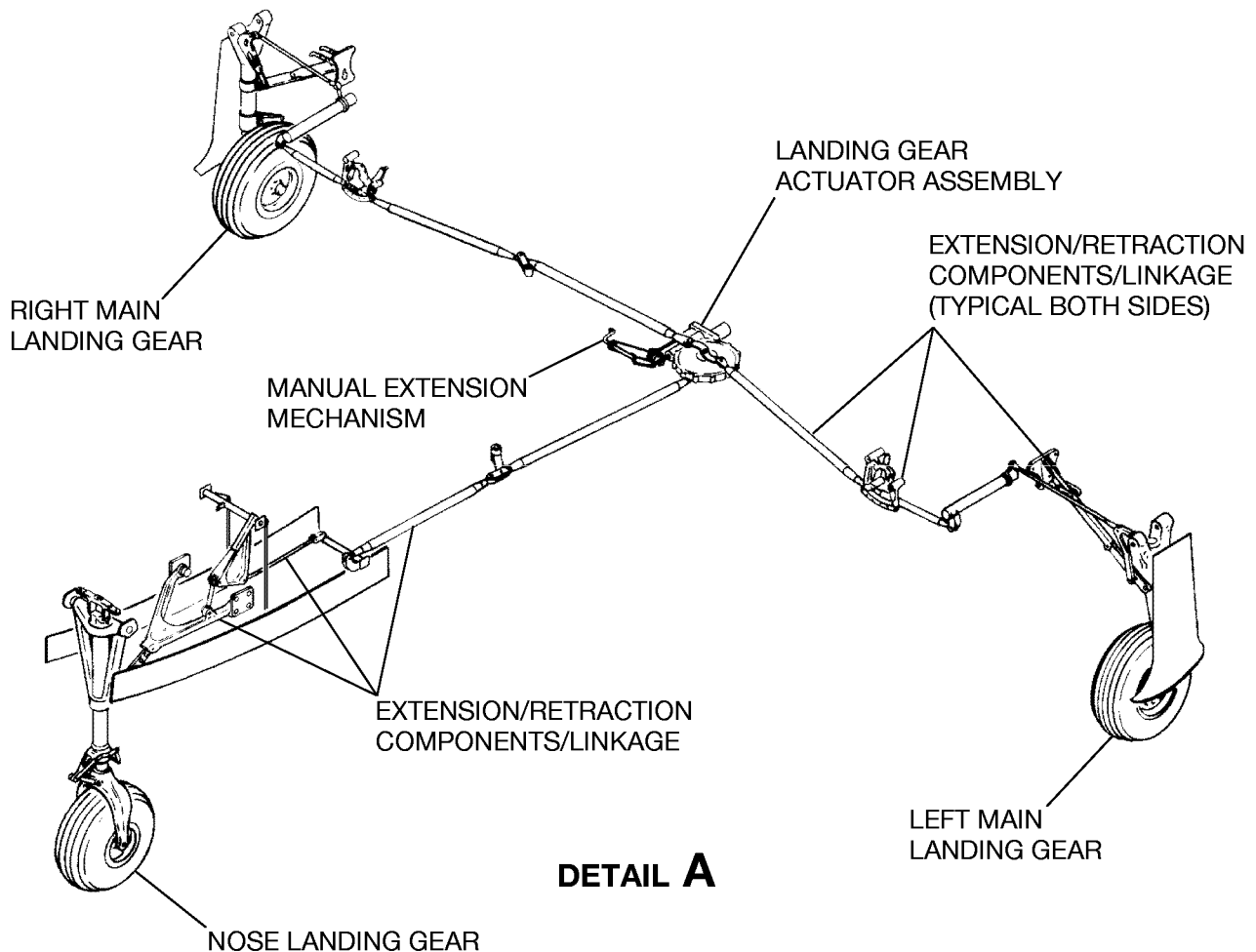
This inspection is intended to assist in preventing gear and gear extension and retraction malfunctions. Refer to MEB88-5, Revision 2, for information on trunnion lug inspection.

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NOTE : REPLACE ALL HARDWARE
REMOVED DURING INSPECTION
PROCEDURE WITH NEW
ATTACHING HARDWARE AT
REASSEMBLY/REINSTALLATION.

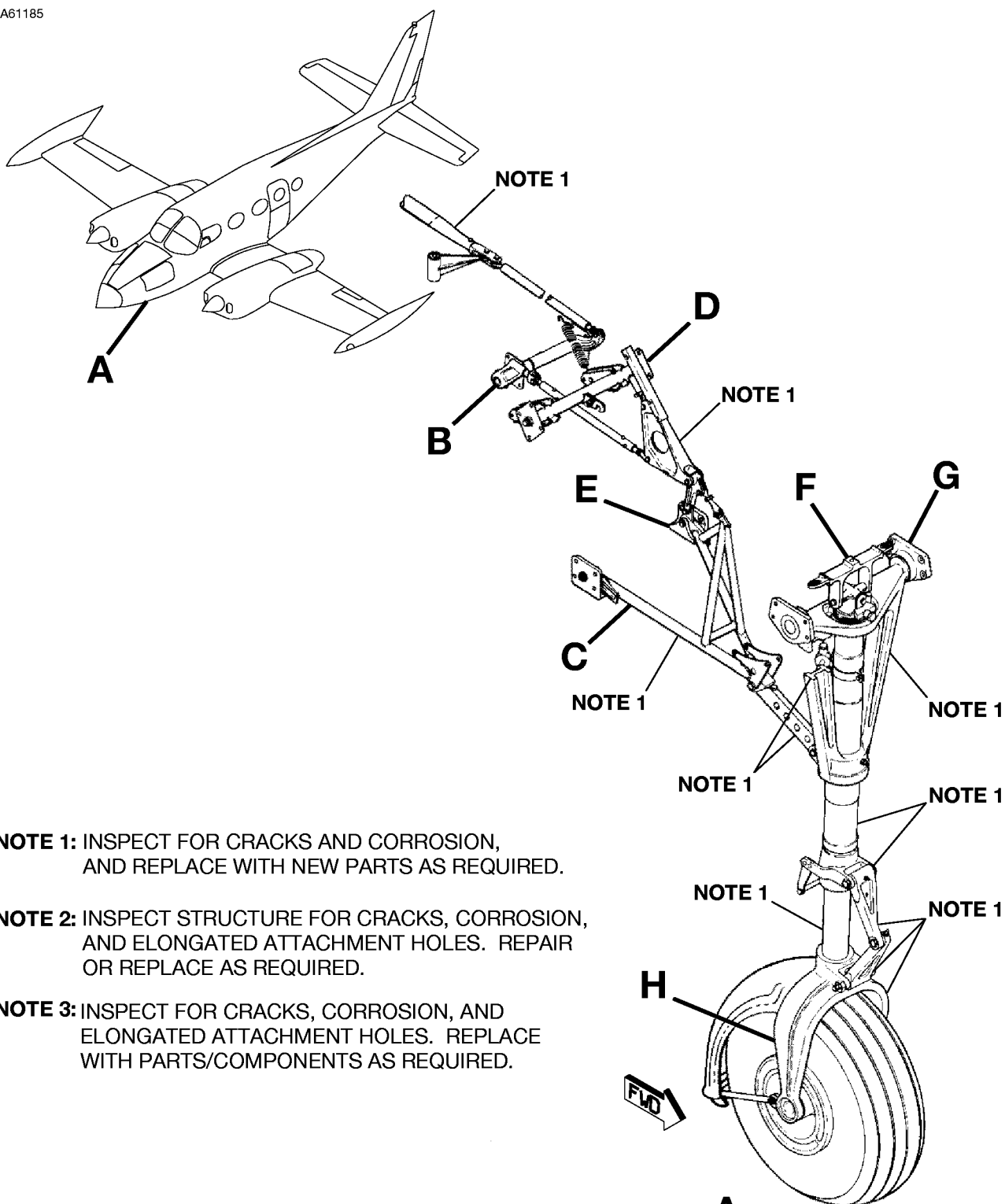


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A1010T1001

Electromechanical Landing Gear System (Typical)
Figure 1 (Sheet 1)

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NOTE 1: INSPECT FOR CRACKS AND CORROSION, AND REPLACE WITH NEW PARTS AS REQUIRED.

NOTE 2: INSPECT STRUCTURE FOR CRACKS, CORROSION, AND ELONGATED ATTACHMENT HOLES. REPAIR OR REPLACE AS REQUIRED.

NOTE 3: INSPECT FOR CRACKS, CORROSION, AND ELONGATED ATTACHMENT HOLES. REPLACE WITH PARTS/COMPONENTS AS REQUIRED.

DETAIL A

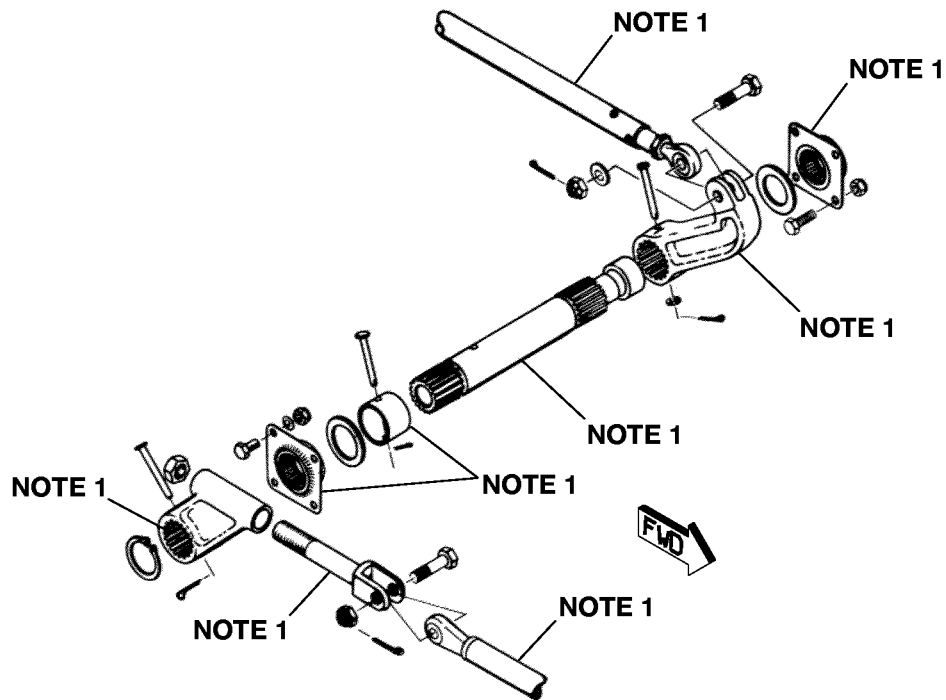
NOSE LANDING GEAR ASSEMBLY
 (AIRPLANES 340-0001 THRU 340-0555)

A1042R3004

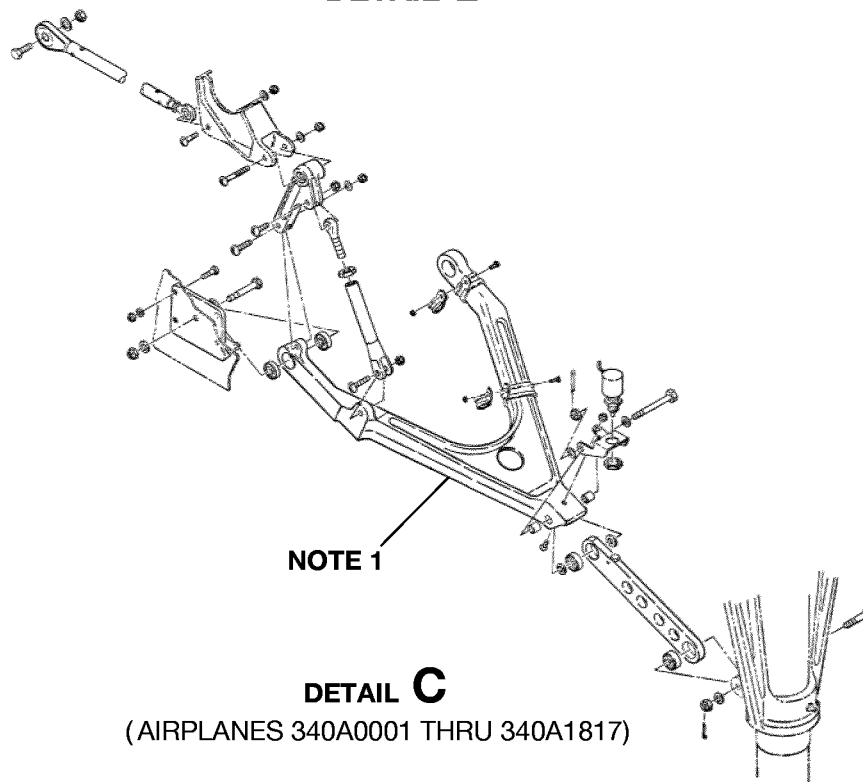
Electromechanical Landing Gear System (Nose Landing Gear)
 Figure 2 (Sheet 1)

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DETAIL B

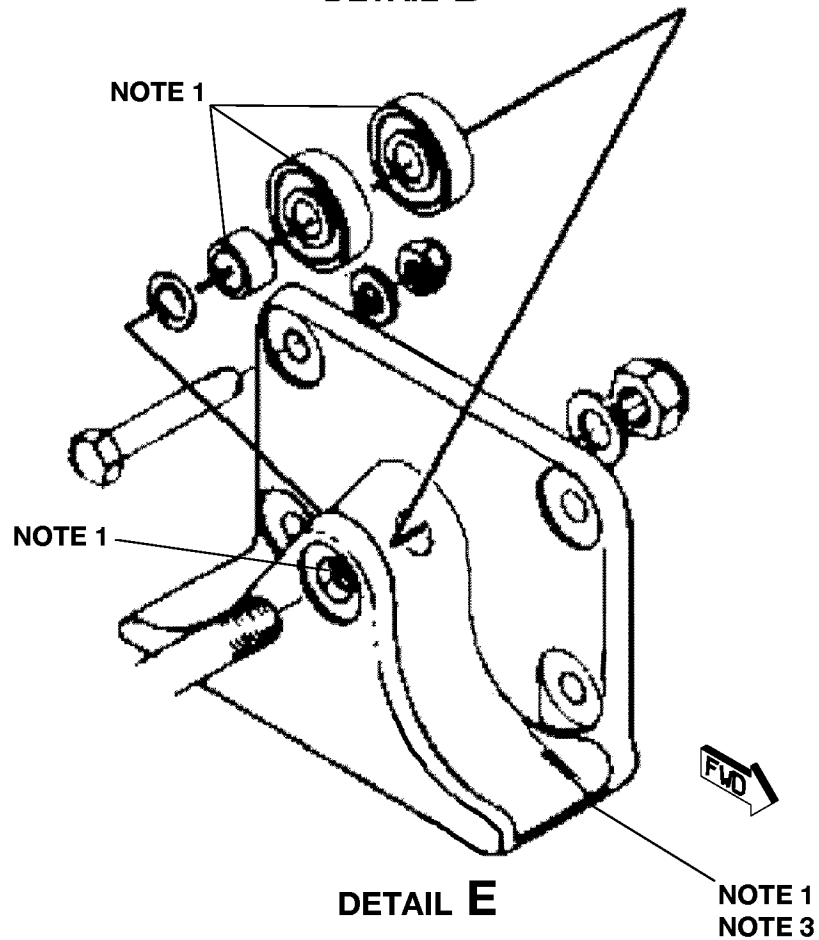
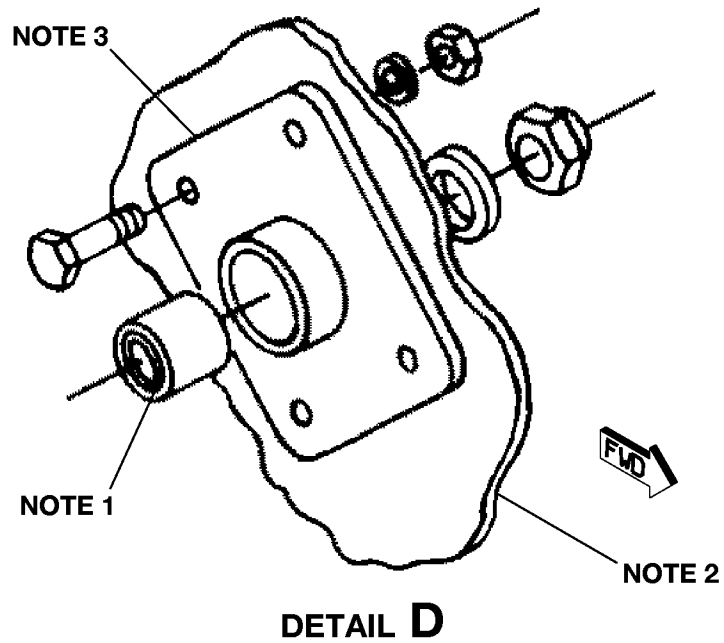


Electromechanical Landing Gear System (Nose Landing Gear)
Figure 2 (Sheet 2)

B1042R2002
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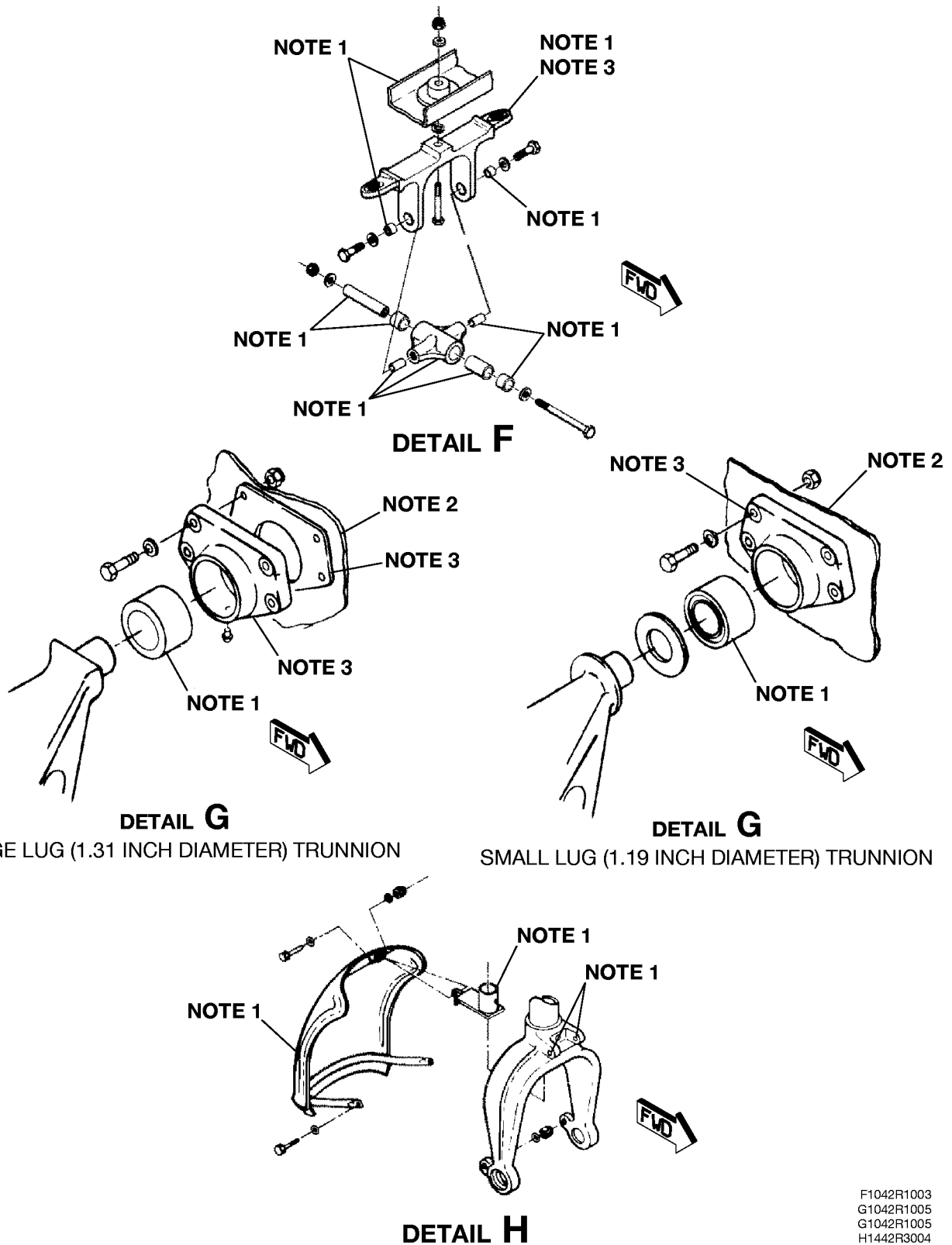


Electromechanical Landing Gear System (Nose Landing Gear)
Figure 2 (Sheet 3)

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DETAIL G
 LARGE LUG (1.31 INCH DIAMETER) TRUNNION

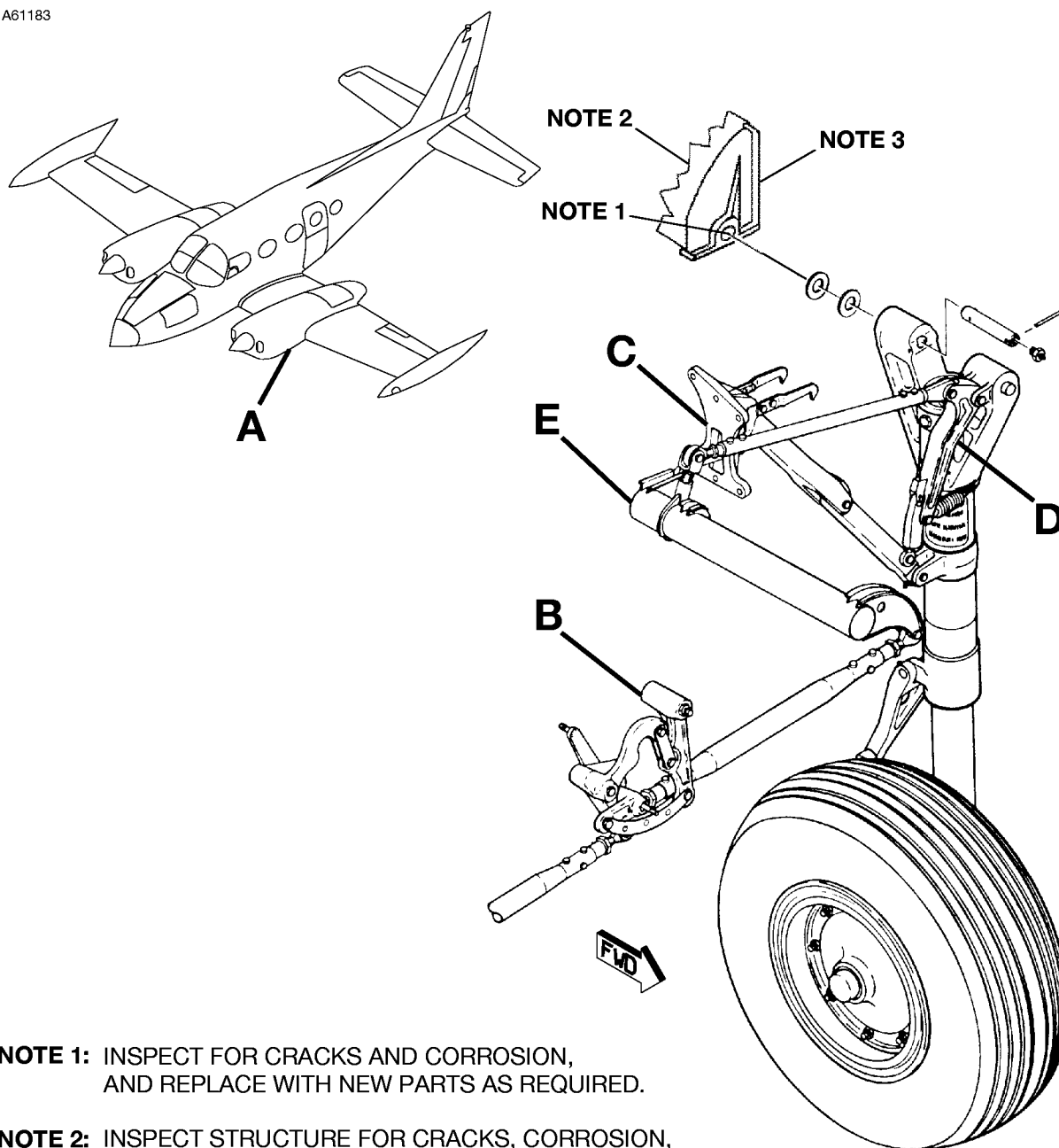
DETAIL G
 SMALL LUG (1.19 INCH DIAMETER) TRUNNION

DETAIL H
 Electromechanical Landing Gear System (Nose Landing Gear)
 Figure 2 (Sheet 4)

F1042R1003
 G1042R1005
 G1042R1005
 H1442R3004

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NOTE 1: INSPECT FOR CRACKS AND CORROSION, AND REPLACE WITH NEW PARTS AS REQUIRED.

NOTE 2: INSPECT STRUCTURE FOR CRACKS, CORROSION, AND ELONGATED ATTACHMENT HOLES. REPAIR OR REPLACE AS REQUIRED.

NOTE 3: INSPECT FOR CRACKS, CORROSION, AND ELONGATED ATTACHMENT HOLES. REPLACE WITH PARTS/COMPONENTS AS REQUIRED.

DETAIL A

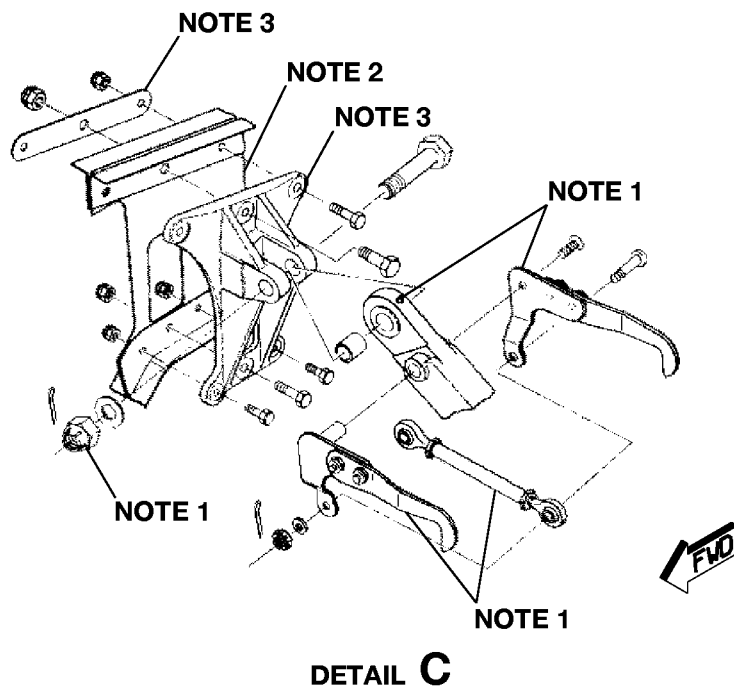
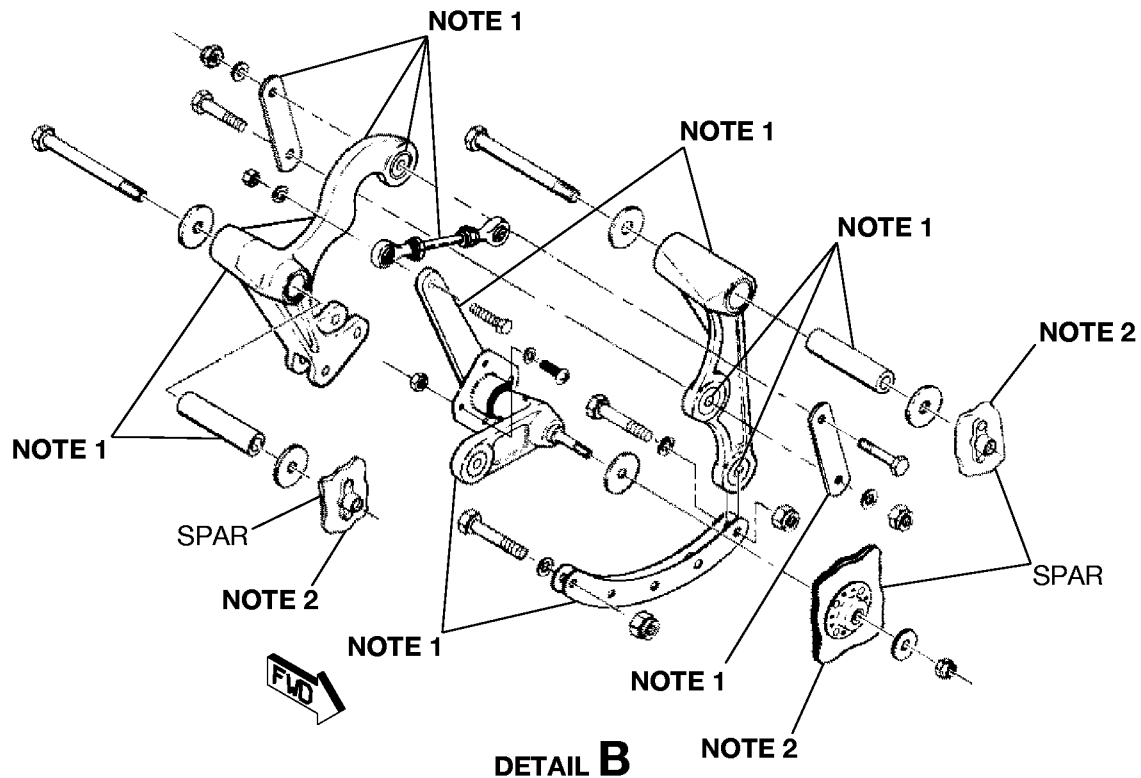
MAIN LANDING GEAR ASSEMBLY
(LEFT SIDE SHOWN, RIGHT SIDE OPPOSITE)

Electromechanical Landing Gear System (Main Landing Gear)
Figure 3 (Sheet 1)

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B5241R1003
C1441R1001

Electromechanical Landing Gear System (Main Landing Gear)
Figure 3 (Sheet 2)

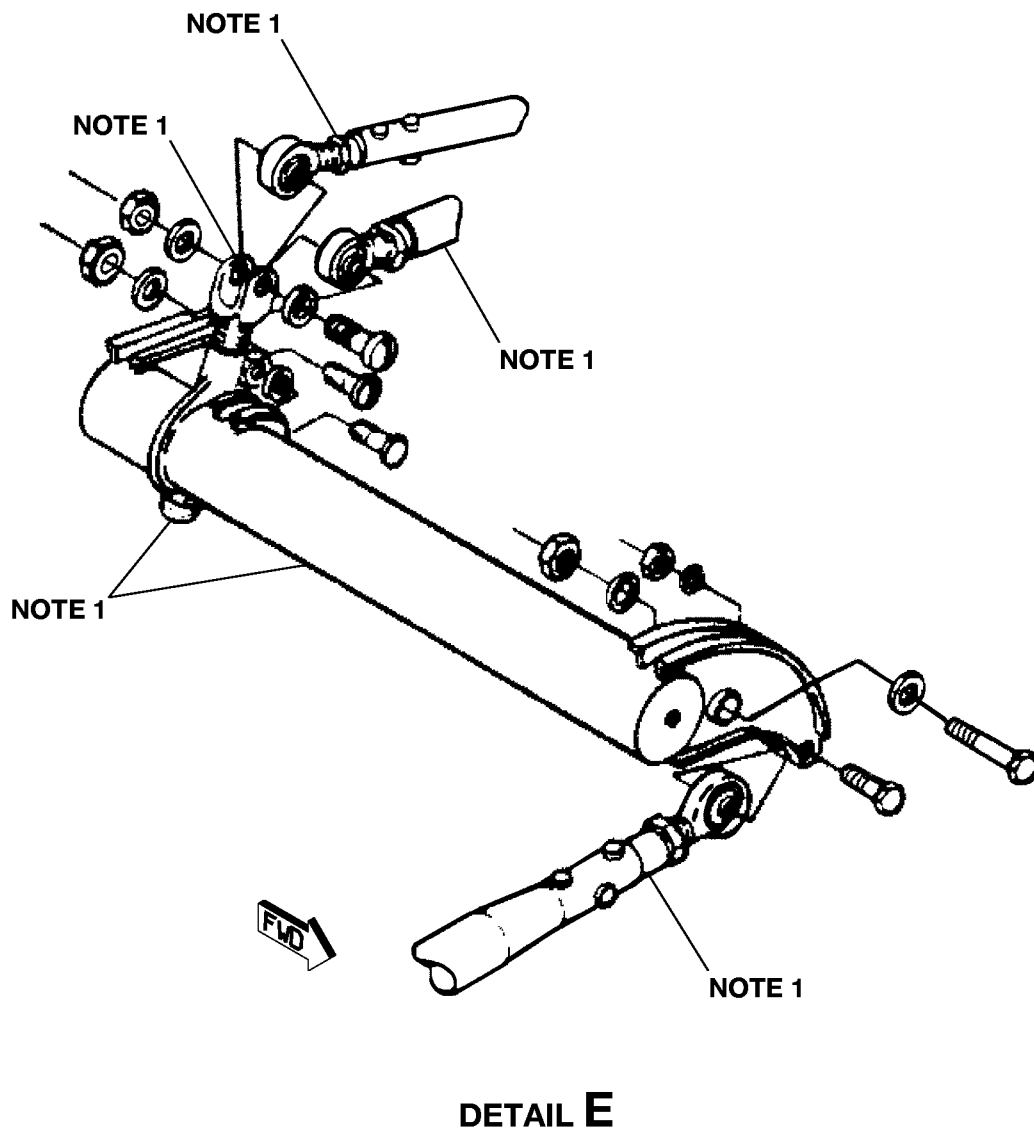
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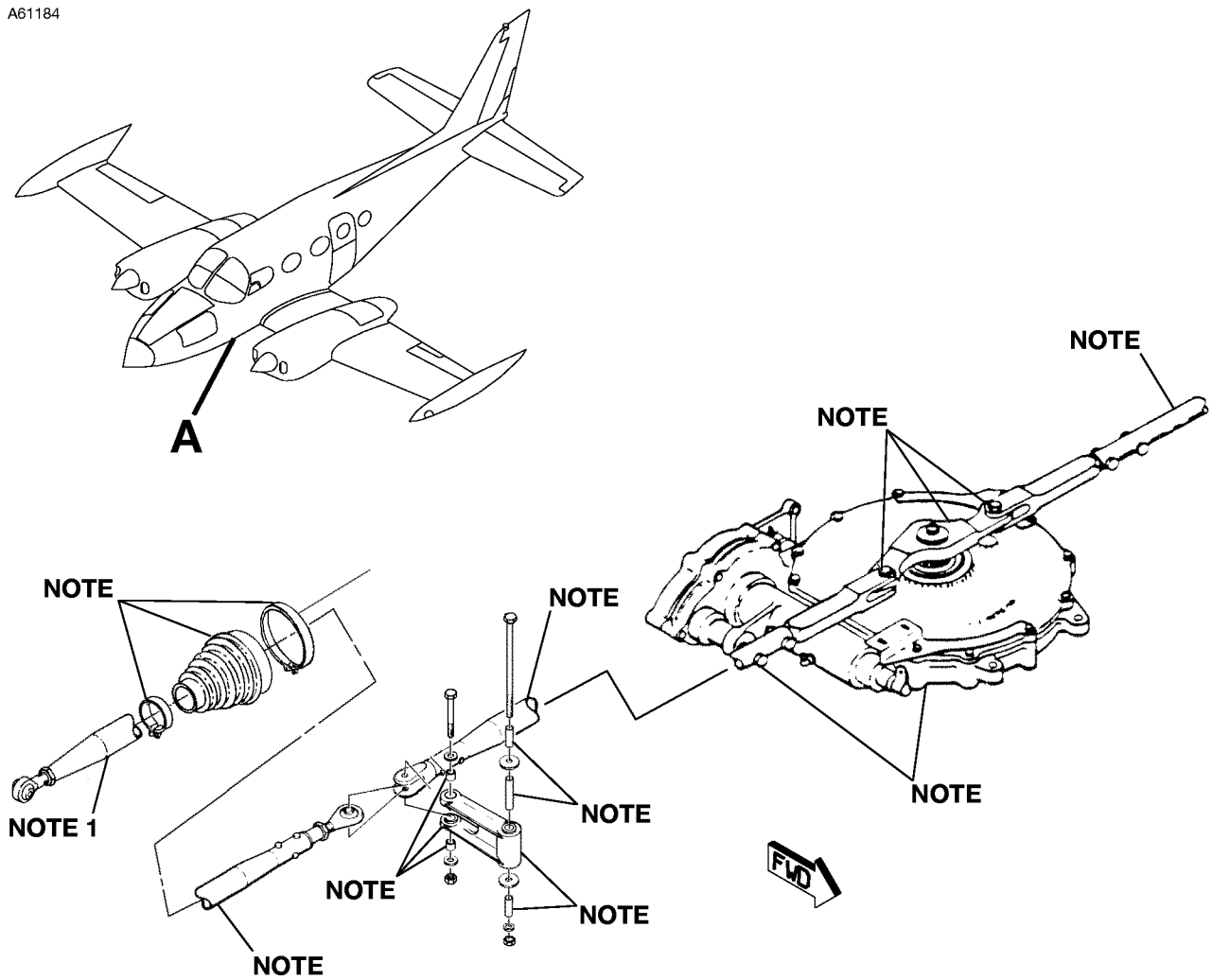


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Electromechanical Landing Gear System (Main Landing Gear)
Figure 3 (Sheet 4)

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DETAIL A
LANDING GEAR ACTUATOR ASSEMBLY

NOTE : INSPECT FOR CRACKS AND CORROSION,
AND REPLACE WITH NEW PARTS AS REQUIRED.

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Electromechanical Landing Gear System (Actuator Assembly)
Figure 4 (Sheet 1)

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SUPPLEMENTAL INSPECTION NUMBER: 32-10-02

1. TITLE

Main Landing Gear Bell Crank Pivot Bolt

2. EFFECTIVITY

340-0001 Thru 340A1817

INSPECTION COMPLIANCE

TYPICAL:	INITIAL	1000 Landings	Or	3 Years
	REPEAT	500 Landings	Or	3 Years

3. PURPOSE

To verify the integrity of the bell crank pivot bolt.

4. INSPECTION INSTRUCTIONS

- A. Remove the pivot bolt in accordance with the Service Manual. Refer to Figure 1.
- B. Inspect the bolt for evidence of shear failure.
- C. Reinstall the bolt if no evidence of shear failure is present. Replace bolt if shear failure is present.

5. ACCESS AND DETECTABLE CRACK SIZE

ACCESS/LOCATION	DETECTABLE CRACK SIZE
Main Landing Gear Wheel Well	N/A

6. INSPECTION METHOD

Visual inspection.

7. REPAIR/MODIFICATION

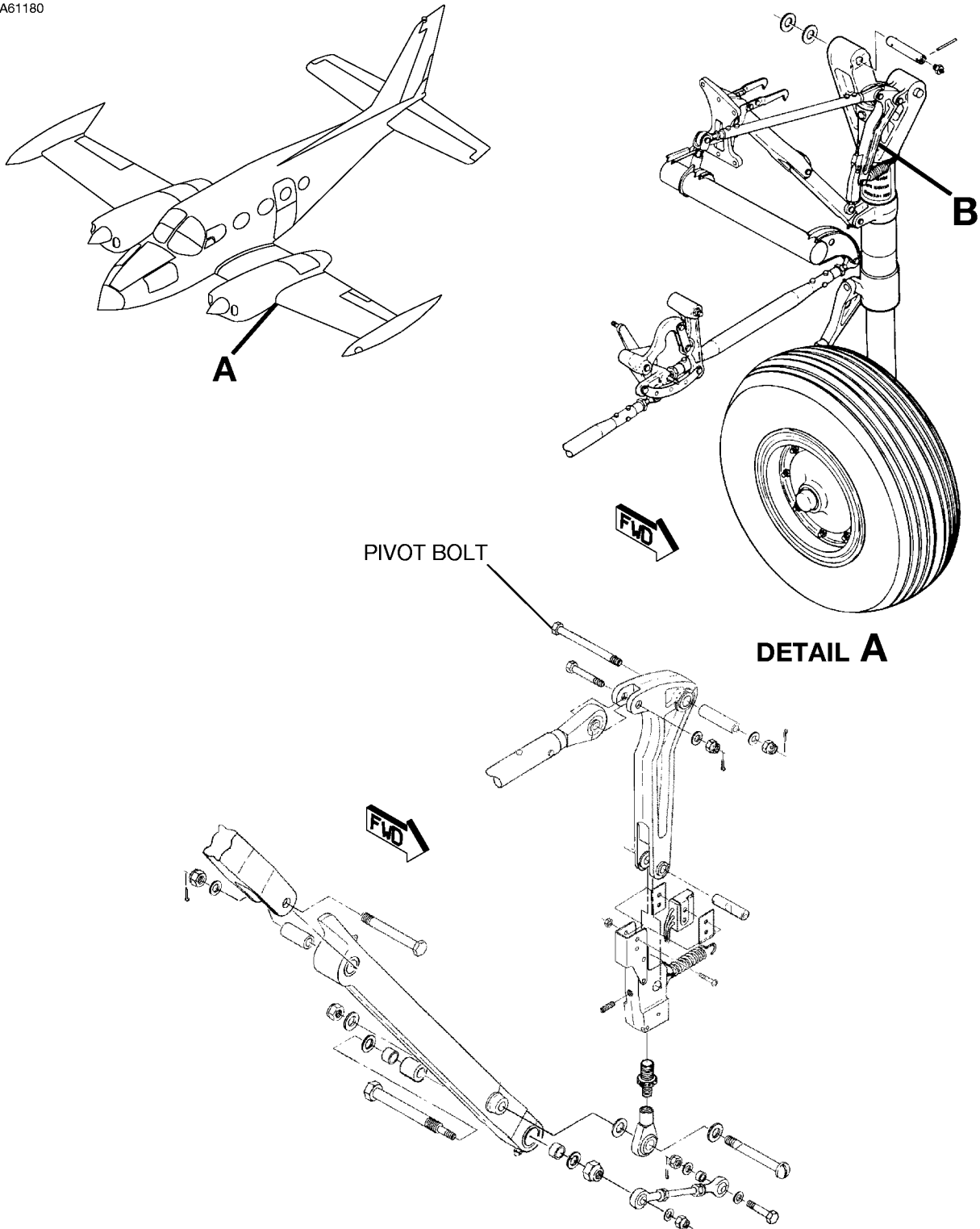
Replace bell crank pivot bolt in accordance with the Service Manual.

8. COMMENTS

Bolt failure will cause the main gear to collapse.

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DETAIL B
(LEFT SIDE SHOWN, RIGHT SIDE OPPOSITE)

Main Landing Gear Bell Crank Pivot Bolt
Figure 1 (Sheet 1)

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